



BLADE STRUCTURE IN A GAS TURBINE

FIELD OF THE INVENTION

This invention relates to a blade structure in a gas turbine. More particularly, this invention relates to a blade structure of a gas turbine with improved turbine efficiency by restricting pressure loss to a minimum level.

BACKGROUND OF THE INVENTION

A gas turbine will be explained with reference to Fig. 16. In general, a gas turbine is equipped with a plurality of stages of stationary blades 2 and 3 arrayed in a circle on a casing (a blade circle or a vehicle chamber) 1, and a plurality of moving blades 5 arrayed in a circle on a rotor (a hub of a base) 4. Fig. 16 shows the moving blade 5 at a certain stage, the stationary blade 2 at the same stage (the inlet side of combustion gas 6) as this moving blade 5, and the stationary blade 3 at the next stage (the outlet side of the combustion gas 6) of this moving blade 5.

When pressure loss is large in the gas turbine, turbine efficiency is lowered. Therefore, it is important to improve the turbine efficiency by minimizing the pressure loss.

However, as shown in Fig. 16, there is a case where the moving blade 5 at a certain stage is what is called a